# Design and Rocket Deployment of a Trackable Pseudo-Lagrangian Drifter based Meteorological Probe into the Lawrence/Linwood EF4 Tornado and Mesocyclone on 28 May 2019

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**Abstract.** A custom lightweight, miniaturized, and trackable meteorological probe was launched by a model rocket into the inflow region ~~of~~ near an EF4, long-tracked tornado south of Lawrence, Kansas, on 28 May 2019 and sampled tornado core flow. The rocket reached apogee at 439 m AGL, releasing the "pseudo-Lagrangian drifter" by parachute directly into the tornado vortex. The probe reached a ~~three-dimensional speed~~ velocity of 85.1 m s$^{-1}$ in the first revolution around the tornado, measured ~~an altitude-corrected~~ a pressure deficit of -113.5 hPa at 475 m MSL, and sampled a tornadic updraft ~~speed~~ of 65.0 m s$^{-1}$. The probe then transitioned to an environment exhibiting ~~a~~ more tilted ascent above an altitude of 4,300 m MSL at speeds up to 84.0 m s$^{-1}$ to a maximum altitude of 11,914 m MSL. 1 Hz pressure, temperature, relative humidity, GPS, acceleration, gyroscope, and magnetometer data for the flight were transmitted in real-time to a ground station until ~~10,680 m MSL and~~ the probe landed 51 km northeast of the launch position. The probe was recovered without damage, which is attributed to the pseudo-Lagrangian Drifter design, and then higher resolution~~and complete~~ 10 Hz data was downloaded~~for the flight~~. This novel deployment method and design facilitate data collection in real-time from within tornadoes, the mesocyclone, and downdraft without requiring the probes to be recovered or for researchers to enter the circulation to deploy equipment.

## 1 Introduction

Supercell tornadoes are ~~among~~ the most damaging local severe weather phenomena capable of producing wind speeds greater than 100 m s$^{-1}$ and atmospheric pressure deficits greater than 100 hPa (Blair et al., 2008; Karstens et al., 2010). The strongest, most damaging winds of a tornado typically occur within the core flow or radius of maximum winds (RMW) along the outer boundary of the tornado core of low atmospheric pressure (Karstens et al., 2010; Wurman et al., 2013). This ring of maximum winds of a tornado is driven by strong horizontal and vertical pressure gradients at the outer periphery of the inner core near and just above the surface (Snow et al., 1980; Lewellen et al., 2000; Fiedler, 2009; Wurman et al., 2013). Direct measurements of above-ground winds and thermodynamics from within the core flow of a tornado had not been accomplished until the present study due to the extreme atmospheric conditions prevalent within the RMW (Wurman et al., 2013)~~and the relative danger of this environment~~.

The authors and others (Snow et al., 1980; Lewellen et al., 1997, 2000; Fiedler, 2009; Karstens et al., 2010; Wurman et al., 2013) theorize that the most significant pressure deficits in tornadoes occur in the core within the RMW very near and just
25  above the ground, often exciting strong, jet-like updrafts that are capable of causing ~~intense, localized~~ damage. High-resolution cameras ~~(Wakimoto et al., 2016) and drones (Groenemeijer, 2022)~~ and drones have documented these damaging updrafts of tornadoes in recent years~~, but direct observations are essential to determine the conditions responsible for the structural damage and human impact. However, these observations are rare and relatively limited.~~ The present study's deployment of miniaturized, pseudo-Lagrangian sensors into tornadoes by rocket can measure the horizontal and vertical winds and pressure
30  perturbations within the tornado's RMW and core.

Direct measurements of atmospheric conditions from inside tornadoes have been limited to ground-based instrumentation (Samaras and Lee, 2004; Blair et al., 2008; Karstens et al., 2010; Kosiba and Wurman, 2013; Wurman et al., 2013). The first ground-based, deployable instrument pack or probe for in situ measurements of meteorological data inside tornadoes was constructed in 1980 and called the TOtable Tornado Observatory (TOTO) (Bedard Jr and Ramzy, 1983; Bluestein,
35  1983, 1999). The usage of TOTO was abandoned when it was determined that placing it directly into the path of a tornado was challenging (Bluestein, 1999).

Tim Samaras recorded a pressure deficit of 100 hPa inside an F4 tornado in Manchester, South Dakota, on 24 June 2003 using the Hardened In Situ Tornado Pressure Recorder (HITPR) (Samaras and Lee, 2004; Karstens et al., 2010). In Tulia, Texas, on 21 April 2007, a mobile mesonet vehicle was struck by an EF2 tornado and recorded a pressure deficit of 194 hPa
40  along with a maximum wind velocity of 50.4 m s$^{-1}$ (Blair et al., 2008). Ground-based measurements of wind and pressure deficit inside a tornado are essential because the surface conditions are responsible for structural damage and human impact. However, ground-based measurements alone do not provide information on the three-dimensional wind, thermodynamics, or structure of tornadoes.

Stirling Colgate of Los Alamos National Laboratory was the first to attempt measurements of meteorological data above-
45  ground and inside tornadoes with instrumented lightweight (<1 lb) rockets launched from a small aircraft (Colgate, 1982). The rocket was fired horizontally at a velocity close to Mach 1 to transect the tornado at altitude while measuring pressure, temperature, ionization, and electric field at 1-meter intervals. During over 120 hours of flying, only one tornado was observed, and the four rocket launches missed the target. Additionally, it was determined that the equipment needed to be hardened and engineering improved. This contrasts with the present study, which uses a rocket launched from the ground as a delivery
50  mechanism for a parachuted, pseudo-Lagrangian sensor.

Over the past ~~three~~ two decades, field experiments such as VORTEX, VORTEX2, and Project TORUS have improved the understanding of supercells, tornadoes, and particularly tornado environments using mobile Doppler radar (X-, W-, Ka-band), in situ ground-based probes, balloons, mobile mesonets, and Unmanned Aerial Systems (UAS) (Straka et al., 1996; Bluestein et al., 2003; Wakimoto et al., 2003; Bluestein et al., 2004; Samaras and Lee, 2004; Blair et al., 2008; Weiss and Schroeder,
55  2008; Karstens et al., 2010; Kosiba and Wurman, 2010, 2013; Wakimoto et al., 2011; Wurman et al., 2012; Tanamachi et al., 2013; Winn et al., 1999; Samaras, 2004; Pazmany et al., 2013; Frew et al., 2020; Houston et al., 2020; Markowski et al., 2018).

Mobile radars are mainly limited to measuring horizontal wind of storms and tornadoes, with the vertical component being inferred due to inclined measurements unless the measurements are taken vertically from dangerous positions inside a tornado. The UAS technology of Project TORUS is not intended for direct measurements inside of a tornado core flow (Frew et al., 2020; Houston et al., 2020).

Multiple-elevation mobile radar data coupled with photogrammetry techniques and or ground-based wind measurements have successfully derived information on the three-dimensional winds of tornadoes (Wakimoto et al., 2011; Kosiba and Wurman, 2010, 2013; Tanamachi et al., 2013). However, mobile radar-based analyses of winds inside tornadoes using multiple elevations are limited in spatial and temporal resolution, range and provide limited information on the thermodynamics of tornadoes (Markowski et al., 2018). Additionally, the UAS technology deployed in the TORUS project is not intended for direct measurements inside of a tornado core flow (Frew et al., 2020; Houston et al., 2020).

A pseudo-Lagrangian drifter is a sufficiently miniaturized, instrumented probe that is intended to move along with the wind flow, such that the speed of the probe is very close to the wind speed of the air in which it is embedded. Using pseudo-Lagrangian drifters for infiltrating thunderstorms and supercells with sensors has been accomplished using a dual-balloon approach (Markowski et al., 2018)and a single-balloon approach where the balloon is filled with helium during flight (Swenson et al., 2019). The thermodynamics of supercell storms (and tornadoes) are essential to understand because wind accelerations and storm morphology are driven primarily by pressure gradient and buoyancy forces (Markowski et al., 2018; Bartos et al., 2022). The wind speeds and accelerations inside a tornado should also be driven by pressure gradient and buoyancy forces. A tornado has yet to be directly sampled by the dual-balloon deployments of Markowski et al. (2018) and Bartos et al. (2022)Markowski et al., 2018; Bartos et al., 2022.

Given the above-mentioned studies, it is clear that there is a notable lack of direct observations inside or near the tornado vortex and mesocyclone. To address this observational need, the present study describes a new, miniaturized, pseudo-Lagrangian sensor deployed from a low-cost surface-based model rocket that is capable of obtaining these observations inside the circulation. The design of this sensor as well as a case study showing its use is presented here. This novel methodology is presented as a solution to measure directly the three-dimensional wind and thermodynamics inside a tornado.

## 2 Methodology

### 1.1 Probe Design

The present study's authors have tested many airborne delivery mechanisms for deploying meteorological sensors into tornado core flows over the past two decades. These include air cannons, parachuted sensor packs dropped by remote-controlled aircraft, and ground-deployed inflatable plastic bubbles to be ingested by the surface inflow of a tornado. After experimenting with many different delivery methods, it was theorized that to obtain meteorological data by a sensor from within a tornado core flow, the speed of model rockets would be required to penetrate subsiding air that is often proximal to a mature tornado (Lemon and Doswell III, 1979). Descending air is often near nearby or even surrounding a supercell tornado because these tornadoes most often occur near strong gradients of vertical velocity (Lemon and Doswell III, 1979).

90 ~~To address the needed observations, the~~ This study presents the design and successful deployment by a rocket of a pseudo-Lagrangian drifter into a tornado vortex and mesocyclone with an overview analysis and discussion of the data received. This novel methodology is presented as a solution to measure directly the three-dimensional wind and thermodynamics inside a tornado.

## 2 Methodology

### 2.1 Probe Design

95 The authors designed and built a custom-engineered probe (Fig. 1) and firmware for the present study. Real-time data reception and location tracking were deemed necessary to ensure that the loss of the probe ~~,~~ due to the hostile environment or inability to recover ~~,~~ did not impact data collection. The software was designed to permit multiple probes to be airborne concurrently.

This lightweight probe has onboard miniaturized sensors for pressure, temperature, humidity, GPS, and a 3-axis Inertial Measurement Unit (IMU). The rocket's pseudo-Lagrangian flight characteristic and small payload capacity necessitated a
100 compact surface-mount design (2.5 x 7.6 cm) with a total mass of 30 g (17 g plus 13 g battery). The probe's onboard sensors were carefully selected based on criteria for wide environmental operating ranges, sampling rates, accuracy, mass, and power efficiency (Table 1). A high sampling rate of 10 Hz was compatible with all sensors, excluding the GPS, which has a maximum rate of 5 Hz.

The GPS sensor selected for the probe was the ORG1411 "Nano Hornet" by OriginGPS. This device has a small form factor
105 and mass while maintaining a GPS fix in accelerations of up to $39.2 \text{ m s}^{-2}$. This sensor has an inbuilt patch antenna which provides weight savings compared to an external antenna.

Similarly, the BME280 sensor by Bosch Sensortec was determined to be an ideal barometric pressure and humidity sensor, given its small mass, size, and performance. The higher pressure sampling rate was advantageous for detecting gravity waves or other fine-scale pressure oscillations within the tornado and parent mesocyclone by comparing GPS altitude to pressure.
110 However, it was determined that the temperature measurement could be improved with the addition of the SI7053 temperature sensor by Silicon Labs. This provided improvements in accuracy and thermodynamic response time ($\tau_{63\%}$) over the BME280 (Table 1).

An IMU was included on the probe to measure the forces and accelerations experienced inside the tornado and the probe's orientation. The 9-axis BNO085 is a miniature IMU by Hillcrest Labs that provides this data from an incorporated 3-axis
115 magnetometer, accelerometer, and gyroscope. The miniature size and mass of the BNO085 are ideal for flight within a tornado. Using quaternions from the IMU and the sensor-fusion, the ~~IMU sensor-fusion of the magnetometer, accelerometer, and gyroscope;~~ Tait-Bryan angles for pitch, yaw, and roll ~~angles~~ (as is used in aircraft orientation) were derived. Information about orientation and forces imparted on the probe help identify turbulence and transitions between shear layers. After manufacturing, a software routine was developed to calibrate the IMU and store the data on the electrically erasable ~~,~~ programmable read-only
120 memory (EEPROM), as stresses built up in the soldering process can offset the factory calibration of micro-electromechanical systems (MEMS) sensors on the die.
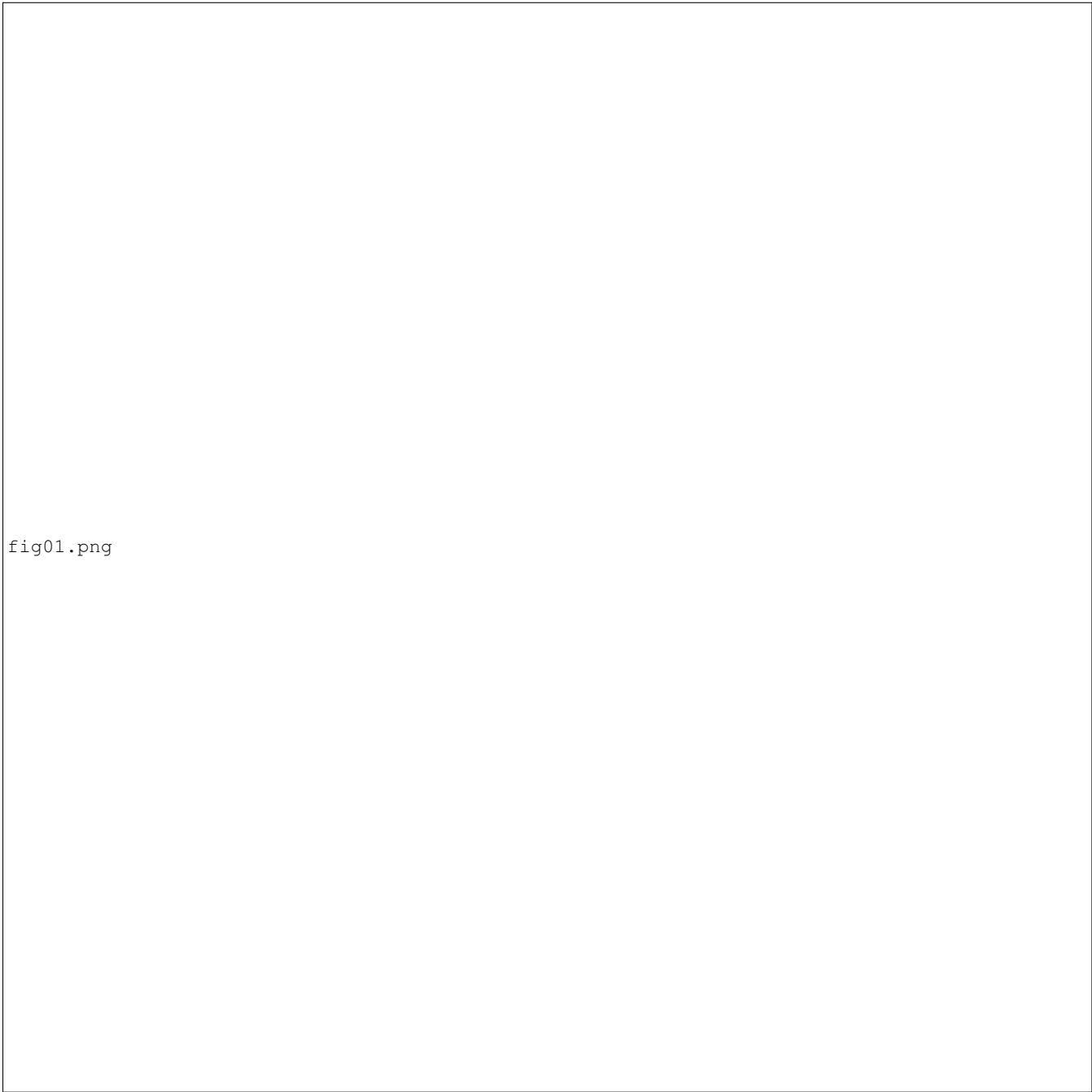
**Figure 1.** 3D render of a custom-engineered probe showing the sensors and major electronic components used in the design.

The probe used Long Range (LoRa) radio transmission to transmit and receive data to and from a ground station. LoRa is a spread spectrum technology that can still work even when the signal is below the noise floor. The power required is

**Table 1.** Probe sensor summary.

| Measurand | Sensor Model | Sampling Rate (Hz) | Mass (g) | Dimensions (cm) | Operating Range (°C) | Specifications |
|---|---|---|---|---|---|---|
| GPS | ORG1411 | 5 | 1.4 | 10.0×10.0×3.8 | -40 to +85 | velocity: <600 m s$^{-1}$<br>acceleration: $\pm$39.2 m s$^{-2}$<br>positional accuracy:<br>  horizontal: <2.0 m<br>  vertical: <3.5 m |
| Pressure, RH | BME280 | 10 | 0.009 | 2.5×2.5×0.9 | -40 to +85 | pressure: $\geq$ 300 hPa<br>RH response($\tau_{63\%}$): 1 s<br>accuracy:<br>  pressure: $\pm$1.7 hPa<br>  RH: $\pm$3% |
| Temperature | SI7053 | 10 | 0.003 | 3.0×3.0×0.7 | -40 to +125 | response($\tau_{63\%}$): 0.7 s<br>accuracy: $\pm$0.3 °C |
| IMU | BNO085 | 10 | 0.043 | 5.2×3.8×1.1 | -40 to +85 | linear acceleration: $\pm$79.5 m s$^{-2}$<br>gyroscope: $\pm$34.9 rad s$^{-1}$<br>magnetometer<br>  $\pm$1,300 $\mu$T (x,y)<br>  $\pm$2,500 $\mu$T (z)<br>accuracy:<br>  lin. acceleration: $\pm$0.35 m s$^{-2}$<br>  gyroscope: $\pm$3.1 ° s$^{-1}$<br>  magnetometer: $\pm$1.4 $\mu$T |

also minimal compared to other technologies, and throughput is traded against range, with longer distances requiring less throughput (Augustin et al., 2016; Noreen et al., 2017). Transmission occurs in the Industrial, Scientific, and Medical band (ISM) at 915 MHz and does not require licensing. Point-to-point LoRa was chosen over the more commonly used wide area star-of-stars topology Long Range Wide Area Network (LoRaWAN) (Semtech, 2019) due to the increased throughput and ability to maintain transmission in sparsely populated areas. A star-of-stars topology is one in which gateways relay messages between devices and a centralized network server.

Time-Division Multiple Access (TDMA) was implemented with regular framing pulses from a ground station, allowing multiple probes to be tracked simultaneously. 1 Hz data transmissions allow the complete sensor data to be transmitted concurrently over LoRa for nine airborne sensors for distances of up to 100 km. The TDMA protocol was modified such that non-reception

of the framing pulse did not result in an immediate loss of synchronization and that opportunistic reception could continue in the event of a loss of synchronization.

135     A 1/4 wave monopole antenna was chosen for the probe and coiled to reduce size. This antenna configuration allowed for an omnidirectional radiation pattern such that the probe's orientation in flight would not impact the communication range.

    The electronics were waterproofed using silicone conformal coating. A permeable and a polytetrafluoroethylene (PTFE) barrier over the port on the BME280

140 to avoid damage to the electronics and unpredictable behavior due to current leakage. Power was provided by a small recharge­able 350 mA·h Lithium-Ion Polymer (LiPo) battery.

## 2.2   Ground Station Design

145 The mobile ground station consisted of a Raspberry Pi Zero W, and a custom LoRa daughterboard with an inter-integrated circuit ($I^2$C) organic light-emitting diode (OLED) display and antenna. A 1/4 wave monopole omnidirectional antenna was used instead of an increased range directional antenna to simplify tracking the probe in a severe weather environment.

    The station contained custom software on an SD Card and provided a web server over a Wi-Fi network. In addition, the station broadcasted the TDMA framing pulse, handled the communication, stored data from the sensors, and provided the user

150 interface for monitoring the sensors in real-time.

    Accessing the interface was via a web browser on a mobile device or computer, and an Apple iPad was used for this purpose. The bearing and distance were calculated from the chase vehicle to the probe by comparing the onboard GPS location of the iPad to the GPS location received from the probe. A 10 A·h rechargeable battery pack provided power.

## 2.3   Rocket and Launcher Design

155 A custom-modified Magnum Sport Loader dual motor model rocket from Quest Aerospace was used to launch the probes into the tornado (Fig. 2). The dual-motor design enhanced thrust, range, and payload capacity.

    High velocity was required to pass through the subsidence often situated nearby an occluded tornado (Lemon and Doswell III, 1979); therefore, higher thrust class D Ammonium Perchlorate motors (Aerotech D21-7T 18mm) were used for each of the

160 dual motors. The motor uses Blue Thunder propellant, has a mass of 25 g, and a peak thrust of 32.1 N. Due to a long 7 s delay between engine burnout and the parachute ejection charge, the rocket coasts further and can cover the distance required to transit into the tornado. The upper three stabilization fins were raised higher on the rocket body to improve stability in high winds. A single hand-sewn nylon parachute with an outside diameter of 0.61 m was used to provide resistance to debris and damage from the ejection charge while lofting in vertical high-velocity winds.

**Figure 2.** Customized rocket (a) and pseudo-Lagrangian drifter deployment configuration (b).

165    Rocket flight simulator software (RockSim Version 9) was used to optimize the center of pressure and achieve maximum flight range without tumbling. Simulations also tested how the rocket would respond to substantial winds and atmospheric

conditions found in the inflow zone of a tornado. The maximum payload capacity of this customized design was determined to be 50 g with a parachute descent rate of approximately 4 - 6 m s$^{-1}$, depending on air density. The probe was housed within the nose cone compartment with expanded polystyrene foam ~~plugs at both ends to constrain the movement of the probe within~~
170 ~~the enclosure during flight and~~ for thermal protection~~of the battery, whilst still permitting airflow across the sensors~~.

A mobile rocket launcher (Fig. ~~4~~2.3) with pan-and-tilt capability was engineered and mounted to the roof of the armored chase vehicle Dominator 3 (Fig. ~~5~~4). The motor and gears of the launcher were derived from the automated window lift system of a junkyard vehicle. The design incorporated a substantial blast plate and two 0.91 m launch rods to launch multiple rockets.

The potential exists for a return lightning strike from a rocket's ionization trail, so a custom control panel was designed
175 inside the Dominator 3 to aim and launch the rockets from within the protective armor (Fig. ~~6~~5).

## 2.4 Launch Safety

The low mass of the rocket, propellant, and payload ~~,~~ combined with the engine size selection results in a Federal Aviation Administration (FAA) Class 1 amateur model rocket designation (Federal Aviation Administration, 2022). Regulations and recommendations for Class 1 rockets in the USA are governed by the FAA, National Fire Protection Association (NFPA), and
180 National Association of Rocketry (NAR). ~~These regulations specify~~

(Federal Aviation Administration, 2022) specifies that a Class 1 amateur rocket must not be launched within a prohibited or restricted area unless permission is obtained, which is determined with aviation maps~~Federal Aviation Administration (2022)~~. Additionally, the rocket must be suborbital, not cross into a foreign nation, be unmanned and "not create a hazard to persons, property, or other aircraft."

185 The NFPA recommendations are adopted at a state level by most states. The minimum site dimension(s) for a Class D motor are 152 m and should be "in a cleared area, free of tall trees, power lines, buildings, and dry brush and grass". A launch of this small size requires a spectator distance of 9 m (National Fire Protection Association, 2018).

NAR summarizes the NFPA recommendations and specifies that the site "must be free of tall trees, power lines, buildings, and dry brush and grass. The launcher can be anywhere on this site, and the site can include roads. Site dimensions are not tied to
190 the expected altitude of the rockets' flights" ~~(National Association of Rocketry, 2022a, b)~~(National Association of Rocketry, 2022b, a).

The risk to aviation is negligible as pilots avoid tornadoes and thunderstorms. The low payload mass of 30 g ~~after deployment~~ is less than a North American Robin, so the impact would be similar to a small bird strike in the improbable event of a collision. After ~~payload separation~~deployment, the probe and parachute behave similarly to other lightweight debris lofted in a tornado ~~at the same~~ and which are at altitude. The descent phase is controlled by a parachute and does not present an impact hazard.
195 ~~Prior to payload separation, the total mass of the rocket, engines, parachute, and payload is 140 g. Comparatively, a typical radiosonde weighs 250 g to 500 g (National Weather Service, 2023) and poses a greater risk to aviation.~~

FAA, NFPA, and NAR have accounted for the risk to people, property, and aviation according to the class definitions they have applied to the hobby and have defined regulations appropriate for safety. Many amateur rocket launches of considerably larger sizes occur daily in the USA, and model rocketry has an excellent safety record. Hitting a vehicle or person is unlikely,
200 and damage at this size and mass is expected to be negligible.

In addition to fulfilling the legal requirements for rocket launches, we also model, sufficiently test rockets, and inspect rockets before launch. The authors keep fire extinguishers available, team members include a firefighter and a fire marshal, and launch site safety is assessed.

## 2.5 ~~Deployment Safety~~

205 ~~The deployment of rocket-borne sensors into tornadoes from the inflow region of supercells is associated with a heightened safety risk, as the path of the tornado often must be traversed to attain an ideal launch position (Fig. 9). The authors followed an intensive safety protocol during this field project, in addition to adhering to all model rocket launch regulations. These safety measures included the use of a custom armored vehicle (Fig. 5) to provide protection from the environment, and a trained spotter in the vehicle for visual observation of the tornado and assessment of its motion prior to deployment. In order~~

210 ~~to maximize safety during field operations, the authors did not deploy from the direct path of the tornado and debris field, and remained inside the vehicle when near the storm to minimize lightning danger. The team extensively rehearsed the deployment procedure prior to this field project, realizing that any inefficiencies or mistakes can have serious consequences near the path of a tornado.~~

## 3 Deployment

215 ### 3.1 Synopsis

The period between 17 May to 30 May 2019 was a prolific period for severe weather across the United States, with 456 tornadoes (NOAA National Centers for Environmental Information - NCEI, 2019b) and three tornado outbreaks (NOAA National Centers for Environmental Information - NCEI, 2019a). Twenty-two were rated EF3, and two were rated EF4, including the Lawrence/Linwood tornado (NOAA National Centers for Environmental Information - NCEI, 2019b).

220 On the morning of 28 May, a robust upper-level trough with 500 hPa temperatures below -20 (°C) covered a large portion of the Mountain West region of North America. A strong southwesterly mid and upper-level jet stream extended downstream over the southern Great Plains. The southerly low-level jet stream (LLJ) of greater than 20 m s$^{-1}$ (40 knots) maintained deep low-level moisture up to a stationary front across northern Kansas and Missouri. An outflow boundary (OFB) from a nocturnal mesoscale convective system (MCS) was draped from northwest to southeast across northeastern Kansas through

225 peak heating (Fig. ~~7~~6) (Storm Prediction Center, 2019).

~~Throughout the day a~~ A dryline would advance east to central Kansas and Oklahoma as an additional focus for supercell development by late afternoon. Beneath the strengthening LLJ, the backed surface winds associated with the OFB in a narrow zone between Kansas City and Topeka would enhance 0-1 km storm-relative helicity to greater than 230 m$^2$ s$^{-2}$; this environment was conducive to violent tornadoes during late afternoon and evening (Fig. ~~7~~6). ~~The NOAA's~~

230 The Storm Prediction Center of NOAA upgraded portions of northeastern Kansas and northern Missouri to a moderate risk area for convective storms in the ~~13:00 UTC~~ 13Z day 1 convective outlook with the potential for strong-to-violent torna-

does (Storm Prediction Center, 2019). An enhanced risk area for convective storms was active to the west and southwest of the moderate risk area for a dryline supercell mode but with lower tornado probabilities given weaker low-level wind shear. Twenty-five tornadoes occurred in the United States on 28 May 2019, of which ten were in Kansas.

## 3.2 Tornado Event

The parent supercell of the Lawrence/Linwood tornado initiated in the open warm sector over the elevated terrain of the Flint Hills at around 20:00 UTC and organized on approach to the OFB located just southwest of Kansas City. The first mesocyclone occlusion yielded a brief EF2 tornado at 22:55 UTC, followed immediately by the rapid development of a more significant, stronger tornado that began in southwestern Douglas County at 23:05 UTC.

The tornado quickly grew to over 1 km wide, producing EF3 damage within ten minutes of tornadogenesis along United States Highway 59 just south of Lawrence. The National Weather Service damage survey confirmed a 51.2 km long track through Douglas and Leavenworth Counties. Peak EF4 damage occurred in Linwood, with maximum winds estimated at 76 m s$^{-1}$ (National Weather Service, 2019) and a damage path width of over 1.6 km (Fig. 87). The tornado dissipated near the town of Bonner Springs, just west of the Kansas City International Airport.

The supercell was of the high-precipitation (HP) class. ~~Previous experience has demonstrated that a~~ The authors had learned from previous experiments that the narrow zone of inflow into the northeast quadrant of a tornado ~~is often~~ would frequently be free of destructive ~~hail which would~~ hydrometeors that could damage the rocket (Fig. 98). The ideal launch location is also just ahead of the wind shift of the RFD gust front to ensure relatively calmer launch conditions (Lemon and Doswell III, 1979; Beck and Weiss, 2013). The easterly and northeasterly inflow winds at this location were more likely to transport the pseudo-Lagrangian drifter into a tornado.

Dominator 3 was positioned at the location of the marked ~~star~~ asterisk (38.898911° N, 95.260437° W, and 258 m MSL) - (Fig. 98), and the rocket launcher aimed into the inflow region. At 23:17:33 UTC, the authors launched the rocket, and the payload was deployed. After launch, the tornado passed south of the launch location, with structural debris visible within the condensation funnel.

## 3.3 Rocket / Probe Flight

At 7.9 seconds after launch, 1 Hz real-time data revealed that the probe parachute deployed at 437 m AGL and entered the northwest side of the tornado. The crew then continued north on Highway 59 in Dominator 3 to maintain communication with the probe as ~~it~~ the probe ascended in the tornado and mesocyclone. At 23:25:09 UTC, communication was lost as the probe approached the tropopause (horizontal distance from the launch of 10.4 km and 10,680 m MSL).

The probe continued the flight and was recovered the next day on the grounds of a church in Leavenworth, KS (39.19° N, 94.94° W), having traveled 51.5 km northeast of the launch location. Subsequently, the authors downloaded the 10 Hz data from the onboard EEPROM for the entire flight.

## 4 Data, Analysis, and Discussion

### 4.1 GPS and Pressure

265 The probe recorded 5 Hz three-dimensional GPS data for time, horizontal position, altitude, speed, and heading. The GPS velocity data recorded by the probe ~~resolves the in situ~~ directly sampled wind speeds during the 30.2-minute flight inside the tornado and parent mesocyclone~~with the assumption that the~~ , which is attributed to the probe being a pseudo-Lagrangian drifter ~~follows the flow with minimal probe-induced motion~~ following the parachute deployment phase.

The GPS trajectory of the probe's flight superimposed on the NWS damage track provides valuable data that could be used

270 for insight into the three-dimensional structure of the violent tornado and its parent mesocyclone (Fig. ~~10~~9). It depicts the rotation of the probe inside the tornado and then entrainment into the tilted updraft of the mesocyclone. ~~Ascent time was rapid with a duration of 10 minutes from launch to the apex of the flight.~~

Owing to the availability of altitude data from the GPS and pressure data from the onboard sensor, it was possible to derive a ~~launch-referenced and altitude-corrected~~ pressure perturbation metric to measure the pressure deficit between the ambient

275 pressure and pressures inside the tornado vortex and mesocyclone (Fig. ~~11~~10e).

The GPS data recorded maxima for the three-dimensional speed of 85.1 m s$^{-1}$ (Fig. ~~11~~10b)~~and~~ , vertical velocity (w)~~of~~ of 65.0 m s$^{-1}$ (Fig. ~~11~~10c)~~at 858 m and 2, 172 m MSL respectively. Additionally, the maximum horizontal speed component of 83.9 m s$^{-1}$ at 856 m MSL (Fig. 11c) was consistent with the three-dimensional speed maxima, indicating that it consisted of mainly a horizontal component. A maximum of~~ , and a maximum altitude of 11,914 m MSL ~~was also recorded~~ (Fig. ~~11~~10a).

280 The GPS Heading showed rotation within the tornado (Fig. ~~11~~10d).

A storm motion of 21 m s$^{-1}$ from 220° was derived from radar and subtracted from the GPS position referenced to the launch location to produce an X/Y position relative to the storm. This was then used to plot storm relative GPS three-dimensional velocity and pressure perturbation (Fig. ~~12~~11). The probe completed 1.5 revolutions around the tornado in near-circular movement with a diameter of 1.6 km (Fig. ~~12~~11a), which is in agreement with the estimated maximum damage width of 1.6 km (National

285 Weather Service, 2019). The 85.1 m s$^{-1}$ wind speed measured by the probe is in accordance with the 76 m s$^{-1}$ wind speed estimated by the damage survey (National Weather Service, 2019), and the pressure perturbation (Fig. ~~12~~11b) indicates a deficit in the range of 80 to 113.5 hPa, which concurs with surface pressure deficits measured in other tornadoes (Samaras and Lee, 2004; Karstens et al., 2010; Blair et al., 2008). Additionally, the wind speed is greater in the southeast portion of the tornado and lower in the northwest, which is attributed to the northeasterly storm motion (Fig. ~~12~~11a).

290 The 39.2 m s$^{-2}$ acceleration limit of the ORG1411 was surpassed for a short period during the powered rocket flight phase, which led to a loss of fix in two-dimensional position for 5 seconds and accuracy in altitude for 60 seconds. However, this still allowed for valuable data to be recorded in the near-ground layer of the tornado~~and the data and figures include this phase of the flight.~~ . The authors have since switched to a GPS receiver with a high dynamics mode that can reacquire a three-dimensional fix after rocket launch within 2 seconds for future deployments.

295 The operating range of the BME280 for barometric pressure extends down to 300 hPa per the Bosch datasheet; however, this flight reached a pressure altitude of 208 hPa. Subsequent cross-analysis between the GPS data and the Topeka 00Z meteorolog-

ical balloon sounding confirmed the validity of barometric pressure readings below 300 hPa ~~due to the closeness of the Topeka sounding combined with minimal and consistent pressure perturbation at the top of the storm and for the descent. It should be noted that during the launch of the rocket prior to pseudo-Lagrangian flight that the pressure was impacted by Bernoulli's~~

300 ~~Principle due to the high velocity of the rocket flight.~~ Improvements to the design since this flight include an additional onboard pressure sensor with an operating range extending down to 100 hPa to serve as a second source of pressure data.

### 4.2 LoRa

During the probe flight, the ground station was handheld ~~external to~~ outside the side window of Dominator 3, and the vehicle resumed highway speeds. Due to debris falling outside the tornado circulation, safety concerns necessitated bringing the ground
305 station back inside the vehicle for periods.

The ground station lost communication with the probe at a range of 14.5 km; however, later recovery of the probe and stored data demonstrated that the probe was still transmitting throughout the entire flight. The authors determined that the shorter-than-expected radio frequency range was due to the proximity of the 16 gauge ballistic steel shell of Dominator 3 and was caused by obstruction of the signal and the Faraday cage behavior.

310 Subsequently, an omnidirectional antenna has been mounted externally on chase vehicles, and the vehicle shell serves as a ground plane. This configuration has demonstrated a line of sight ground level range of 26 km and an airborne range of over 100 km.

### 4.3 Temperature and Relative Humidity

Holes were punched into the nose cone compartment of the rocket to promote airflow across the temperature and relative
315 humidity sensors during flight, as pseudo-Lagrangian drifting within the tornado limits the airflow ~~relative to the probe~~. Additionally, with minimal ~~probe-relative~~ airflow, the effect of the electrical self-heating from the radio transceiver needed to be modeled. Other considerations were that the temperature within Dominator 3 before launch was higher than the external temperature and the probe's thermal mass was moved from inside the vehicle to outside for launch.

Due to operational and time constraints prior to launch, a thermal analysis of the enclosure was completed after recovery.
320 Treatments for thermal response exist for radiosondes that exhibit similar concerns where slower ascent rates result in a lack of airflow past the sensors contained within (Mahesh et al., 1997; Miloshevich et al., 2004).

Controlled experiments (Fig. ~~13~~12) were conducted to compute the thermal time constant ($\tau$) of the probe within the enclosure, the probe outside the enclosure, and the self-heating effect ~~(Simpson and Timmer, 2019)~~. A fan, handheld anemometer, deep freezer, and ice-bath calibrated thermocouple were used.

325 These experiments identified constants for thermal response ($\tau$) and self-heating (S) to correct the temperature profile as measured by the probe. Additionally, the exact temperature was known at two points in time; launch (due to the calibrated weather station on the roof of Dominator 3) and at a point in the descent where the air temperature external to the enclosure of the probe was equal to the measured temperature within the enclosure of the probe, and therefore no heat transfer was occurring ~~to or from the sensor internal to the probe (Fig. 14).~~.

330 Equation (1) was derived to correct the thermal data based on Newton's Law of Cooling (Miloshevich et al., 2004; Mahesh et al., 1997; Nash, 2015; World Meteorological Organization, 2017). ~~Although the descent rate of the payload was known when moving~~

Air flow around the probe (v) was unknown but near zero as the payload moved as a pseudo-Lagrangian drifter~~, the unknown airflow around the sensor (v) could be determined to lie within a range of possible values. The sensor airflow (v) must be less~~

335 ~~than the descent rate due to the enclosure having venting but not being completely open to the environment.~~ A meteorological balloon sounding (Topeka 00Z) was used as a reference for the environmental temperature profile by mapping the interpolated altitude in the sounding to the GPS altitude recorded by the sensor. It was found that there was a minimal range for plausible values for (v) that did not cause the corrected temperature to overshoot (v=0.07 m s$^{-1}$) the Topeka profile or undershoot (v=0.27 m s$^{-1}$). A midpoint value for (v) was chosen of 0.17 m s$^{-1}$ (Fig. ~~14~~13), which provided the best curve fit

340 between the corrected temperature and Topeka profiles.

$$T_{env} = \frac{(T_t - S) - (T_0 - S)\,e^{-\Delta t/\tau}}{1 - e^{-\Delta t/\tau}} \tag{1}$$

where:

$\tau$ = time constant [Eq. (2)] ),    $T_t$ = sensor temperature at time t ($^\circ$C),

$T_{env}$ = temperature of the environment ($^\circ$C),    $T_0$ = sensor temperature at time 0 ($^\circ$C),

345 $\Delta t$ = delta time $t_t - t_0$ (seconds),    $S$ = self heating (4.18 $^\circ$C)

$$\tau = \tau_0\,(\rho v)^{-n} \tag{2}$$

where:

$\tau_0$ = time constant at zero air velocity (seconds),    $\rho$ = air density [Eq. (3)] ,

$v$ = air velocity in relation to the sensor (m s$^{-1}$),

350 $n$ = constant ranging from 0.4 (laminar air) to 0.8 (turbulent air); 0.8 is used

$$\rho = \frac{p}{RT} \tag{3}$$

where:

$p$ = sensor pressure (pascals),    $R$ = 287.058 (gas constant for dry air),

$T$ = 00Z Topeka sounding temperature ($^\circ$K)

355

Due to the complexities and cost (> \$15,000) of the experimental setup to categorize the response for relative humidity in the enclosure, the authors decided to forego response time correction for this data; consequently, this necessitated using temperature instead of virtual temperature for the ideal gas equation [Eq. (3)]. This impact was deemed negligible for this

study as condensing moisture was not encountered by the sensor, and design improvements had already been completed for

360     future deployments negating the requirement for thermal response correction. Additionally, this data component could be categorized later if required and budget permitted.

    The design has since been improved to remove the requirement for the above data treatment by mounting the temperature and relative humidity sensors on a small thermally isolated daughterboard on the exterior of the rocket nose cone.

    The effect of direct solar radiation (Wang et al., 2013) on the thermal correction was considered; however, as most of the

365     flight was within cloud and the sensor was shielded from direct solar radiation by the enclosure, it was concluded that any impacts from this would be negligible. To address this, future design improvements use two identical external temperature sensors, one black and one gold.

## 5   Summary and Conclusions

The authors launched a lightweight real-time trackable meteorological probe by model rocket into an EF4 tornado. Kinematic

370     and thermodynamic data were obtained from inside a tornado and associated mesocyclone. The high velocity of the model rocket successfully passed through any sinking air between the launch position and the tornado, and the probe was deployed by the parachute and pulled inside the tornado. Three-dimensional direct measurements of wind, pressure, and temperature within a tornado, including updraft, had yet to be accomplished until the present study.

    The ground-based launch of a rocket into the inflow region of a supercell proved effective in deploying a parachuted probe

375     into the tornado while avoiding the largest hydrometeors. The pseudo-Lagrangian trajectory of the probe allowed for the recording of three-dimensional wind speed through high-resolution 5 Hz GPS data.

    While thermodynamic gradients are difficult to diagnose from the data of a single pseudo-Lagrangian trajectory, the success of the present study in capturing direct measurements of wind, pressure, and temperature inside a tornado will set the stage for future deployments of multiple simultaneous probes.

380     Multiple pseudo-Lagrangian probes collecting high-resolution meteorological data inside a tornado would allow a better understanding of pressure gradient and buoyancy forces. These forces drive the winds and behavior of tornadoes and supercells (Markowski

    The probe transmitted high-resolution 1 Hz data in real-time from inside the tornado and provided 10 Hz (5 Hz GPS) data upon recovery. A flight time of 30 minutes sampled high three-dimensional wind speeds of up to 85.1 m s$^{-1}$, an updraft

385     velocity of 65.0 m s$^{-1}$, and reached the upper troposphere at 11,914 m MSL before descending. The probe completed 1.5 revolutions around the tornado in near axisymmetric flow during the 3 minutes within the tornado with a maximum updraft velocity of 65.0 m s$^{-1}$. The substantial pressure deficit during the tornado phase of the probe trajectory and the lack-there-of during the mesocyclone is expected due to tornadic winds being driven by tornado-scale gradients in pressure.

    The 65.0 m s$^{-1}$ updraft velocity is the first direct measurement of vertical wind from inside a tornado. A maximum pressure

390     deficit ~~of 113.5~~ of -113.5 hPa was recorded at an altitude of 475 m MSL within the tornado and dropped to less than -20 hPa by an altitude of 3,760 m MSL in the mesocyclone. The most significant pressure deficit being located just above the ground

in this tornado would excite strong vertical velocities like those measured (65.0 m s$^{-1}$). Additionally, the maximum GPS three-dimensional speed of 85.1 m s$^{-1}$ at an altitude of 858 m MSL concurs with the surface wind of 76 m s$^{-1}$ estimated by the NWS damage survey; the trajectory within the tornado was near-circular, and the diameter of the probe's flight was in agreement with the damage survey of 1.6 km.

The sharp transition of the probe trajectory over 10 s from a closed rotational path to pure updraft with negligible curvature starting at 23:19:40 UTC and an altitude of 1,128.7 m MSL (Fig. ??14F), along with a significant pressure deficit of -113.3 hPa (Fig. ??14G), is supportive of the probe sampling the RMW and the core of the tornado. The cessation of the rotational path of the probe is determined by the termination of the 90° phase difference between the u and v GPS sinusoidal velocity components that indicate circular rotation together with a sudden increase in the updraft speed. The pressure deficit was still significant (-66.7 hPa) when the maximum updraft of 65.0 m s$^{-1}$ was recorded at an altitude of 2,171.3 m MSL, with near-zero curvature evident in the probe trajectory.

Previous studies involving radar observations, real ground measurements of wind and pressure, laboratory experiments, and computer simulations have shown that the inner core of a tornado inside of the core flow/RMW is comprised of a consistently large pressure deficit with relatively minimal pressure variation within the core (Snow et al., 1980; Lewellen et al., 1997, 2000; Karstens et al., 2010; Wurman et al., 2013). The high-resolution modeling of tornadoes (Lewellen et al., 1997, 2000) shows a relatively consistent pressure deficit within the tornado core for both one and two-celled vortices. A secondary maximum in pressure deficit as the rotational path of the probe abruptly ended is supportive of the transition of the probe from the RMW to the tornado's inner core. Therefore, the miniaturized pseudo-Lagrangian drifter effectively sampled meteorological data within the core flow and into the low-pressure core of a strong tornado, despite the extreme conditions and flying debris that typify core flow.

After 1.5 revolutions around the tornado, the probe underwent a consistently tilted, ascending path from an altitude of 4,300 m MSL up to the apex of 11,914 m MSL with a maximum GPS velocity of 83 m s$^{-1}$. The high ascent velocity, combined with the trajectory's lack of rotation and relative linearity, indicates that the probe was entrained very close to the center of the mesocyclonic updraft. The authors reached this conclusion as a rotating mesocyclone flow was not indicated in the data as evidenced by the GPS heading and, therefore, lack of probe rotation around a center.

## 6    Lessons Learned / Future Work

The authors have since switched to a GPS receiver with a high dynamics mode for future deployments to reduce the reacquisition time for the GPS position immediately after the rocket launch. This improvement enables the acquisition of a three-dimensional fix within 2 seconds following the rocket launch.

The design has since been improved to eliminate the requirement for the thermal response data treatment by mounting the Improvements to the design since this experiment have included temperature and relative humidity sensors on a small thermally isolated daughterboard on the exterior of the external to the rocket nose cone. This approach provides similar thermal response times to the specifications for these sensors. Additionally, the design has been modified to include an additional onboard

pressure sensor with an operating range extending down to 100 hPa to serve as a second source of pressure data that is specified into the stratosphere.

Although direct solar radiation did not impact the present study, two identical external temperature sensors, one black, and one gold, have been added so that the probe can also be used for balloon deployments. An , a high dynamics GPS, and an omnidirectional rooftop antenna is now used on the chase vehicle to improve radio reception distance, with the metal roof of the vehicle serving as a ground plane. This configuration has demonstrated a line of sight range of 26 km horizontally and an airborne range of over 100 km.

While thermodynamic gradients are difficult to diagnose from the data of a single pseudo-Lagrangian trajectory, the success of the present study in capturing direct measurements of wind, pressure, and temperature inside a tornado will set the stage for future deployments of multiple simultaneous probes and . Future work related to this project will involve simultaneously launching multiple probes from multiple chase vehicles. The authors believe this will provide more insight into the three-dimensional structure and processes within tornadoes.

Multiple pseudo-Lagrangian probes collecting high-resolution meteorological data inside a tornado would allow a better understanding of pressure gradient and buoyancy forces. These forces drive the winds and behavior of tornadoes and supercells (Markowski

*Data availability.* The data received from the probe has been made available under a Creative Commons CC-By Attribution 4.0 International license: https://www.doi.org/10.17605/OSF.IO/Z64FD (Timmer et al., 2022)

*Video supplement.* A video animation of the probe trajectory (Supplemental1.mov) and a video recording of the rocket launch (Supplemental2.mov) has been made available under a Creative Commons CC-By Attribution 4.0 International license: https://www.doi.org/10.17605/OSF.IO/UBTN5

*Author contributions.* RT conceptualized the project, managed the field work, and produced the rough draft of the manuscript; MS designed and built the probes, wrote the firmware, software, tested and curated and analyzed the data; MS prepared the manuscript in conjunction with RT and contributions from CB; CB designed and constructed the rockets, launch system, and maintained Dominator 3; SS drove Dominator 3 and provided safety monitoring.

*Competing interests.* The authors declare that they have no conflict of interest.

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3D render of rooftop rocket launcher (credit: OUTBRK). Rockets and blast plate (a); weather station and rocket launcher pan/tilt mechanism (b).

3D render of rooftop rocket launcher (credit: OUTBRK). Rockets and blast plate (a); weather station and rocket launcher pan/tilt mechanism (b).

**Figure 3.** ~~Illustration of the stages of the rocket flight.~~

3D render of rooftop rocket launcher (credit: OUTBRK). Rockets and blast plate (a); weather station and rocket launcher pan/tilt mechanism (b).

**Figure 4.** Dominator 3 storm chasing vehicle (credit: OUTBRK).

fig05.jpg

**Figure 5.** Custom control panel for launching rockets.

**Figure 6.** Meteorological synopsis for the May 28 May 2019 tornado event.

**Figure 7.** Damage and intensity track of Lawrence/Linwood tornado (credit: NWS).

**Figure 8.** NWS Topeka radar (KTWX) 23:12 UTC; elevation 0.7°; county and state boundaries, rivers, and major roads are depicted; reflectivity and velocity in bottom-right (Lemon and Doswell III, 1979; Beck and Weiss, 2013) (credit: RadarScope/DTN).

**Figure 9.** Probe GPS trajectory ~~retrieved from the 10Hz data combined~~ with NWS damage track ~~are~~ superimposed ~~on top of a Google Earth map. The section of flight within the~~ and tornado ~~is displayed as a close-up~~ close up (left panel). (© Google Earth).
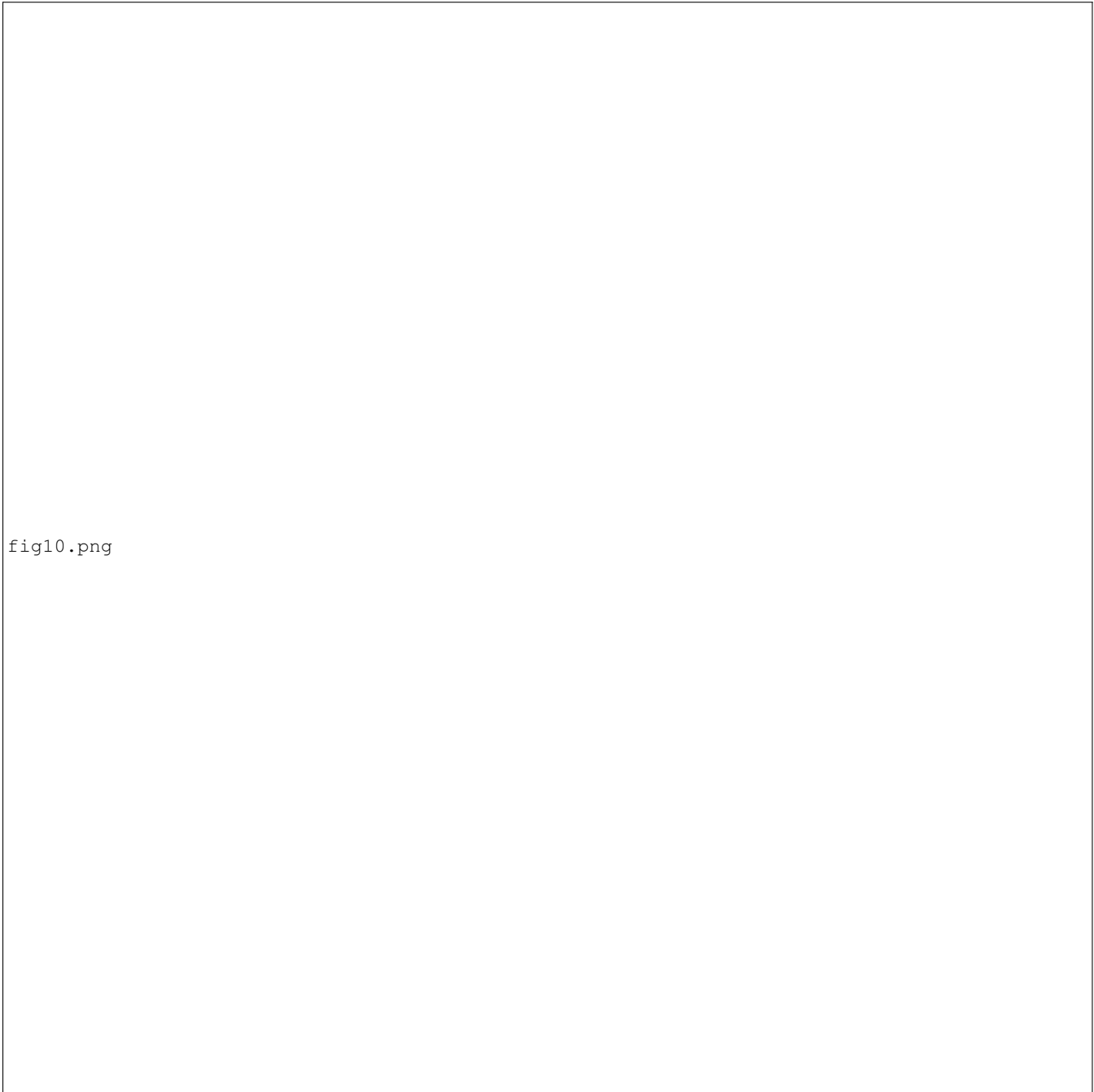
**Figure 10.** GPS data and pressure perturbation sensor.~~The dotted lines separate tornado and mesocyclone flow from the remainder of the flight.~~
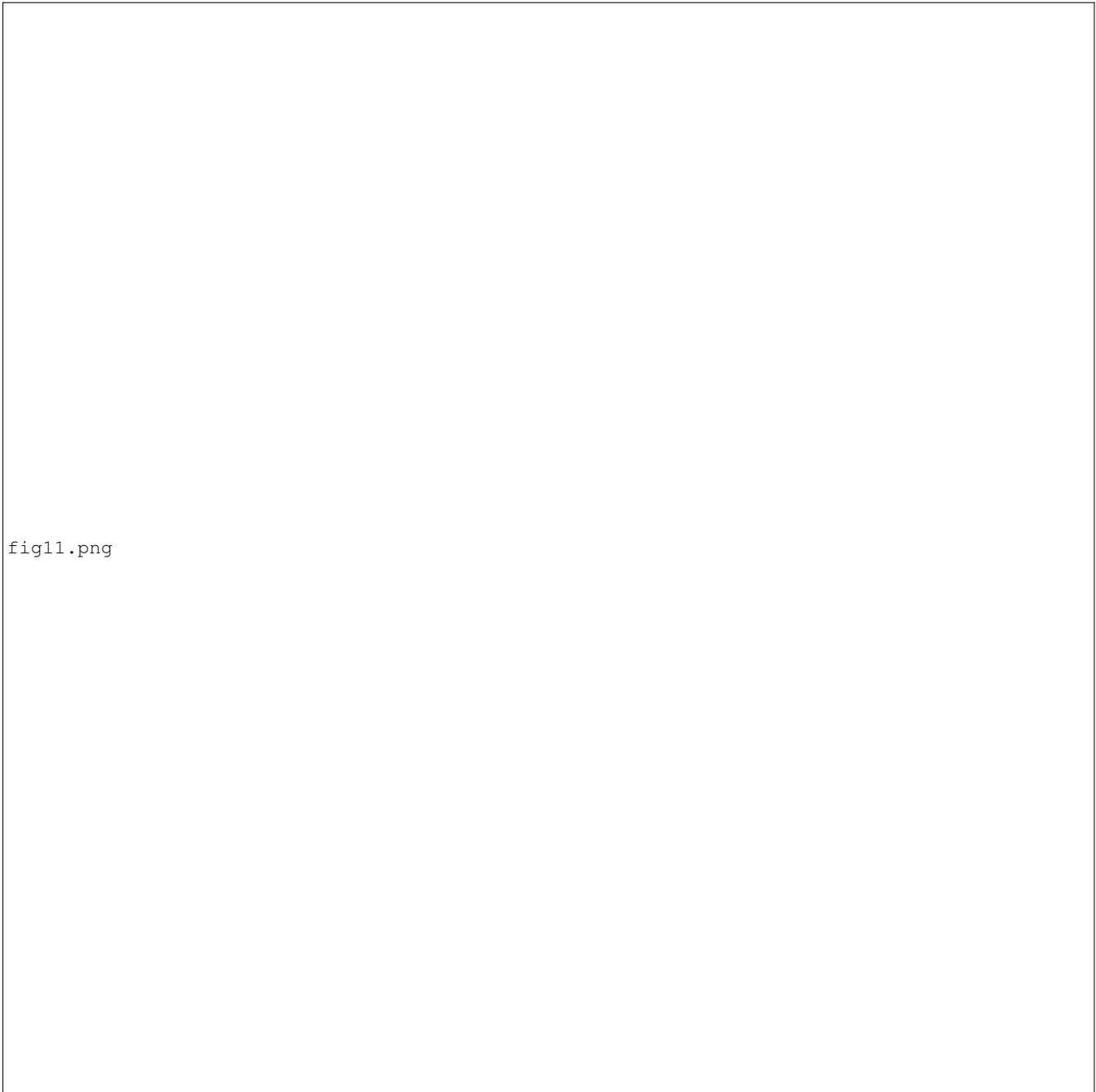
**Figure 11.** Storm relative probe position for the flight with GPS 3-D Speed (a) and Pressure Perturbation (b).

**Figure 12.** Measuring the thermal time constant after removal from a deep freezer with a linear fan.
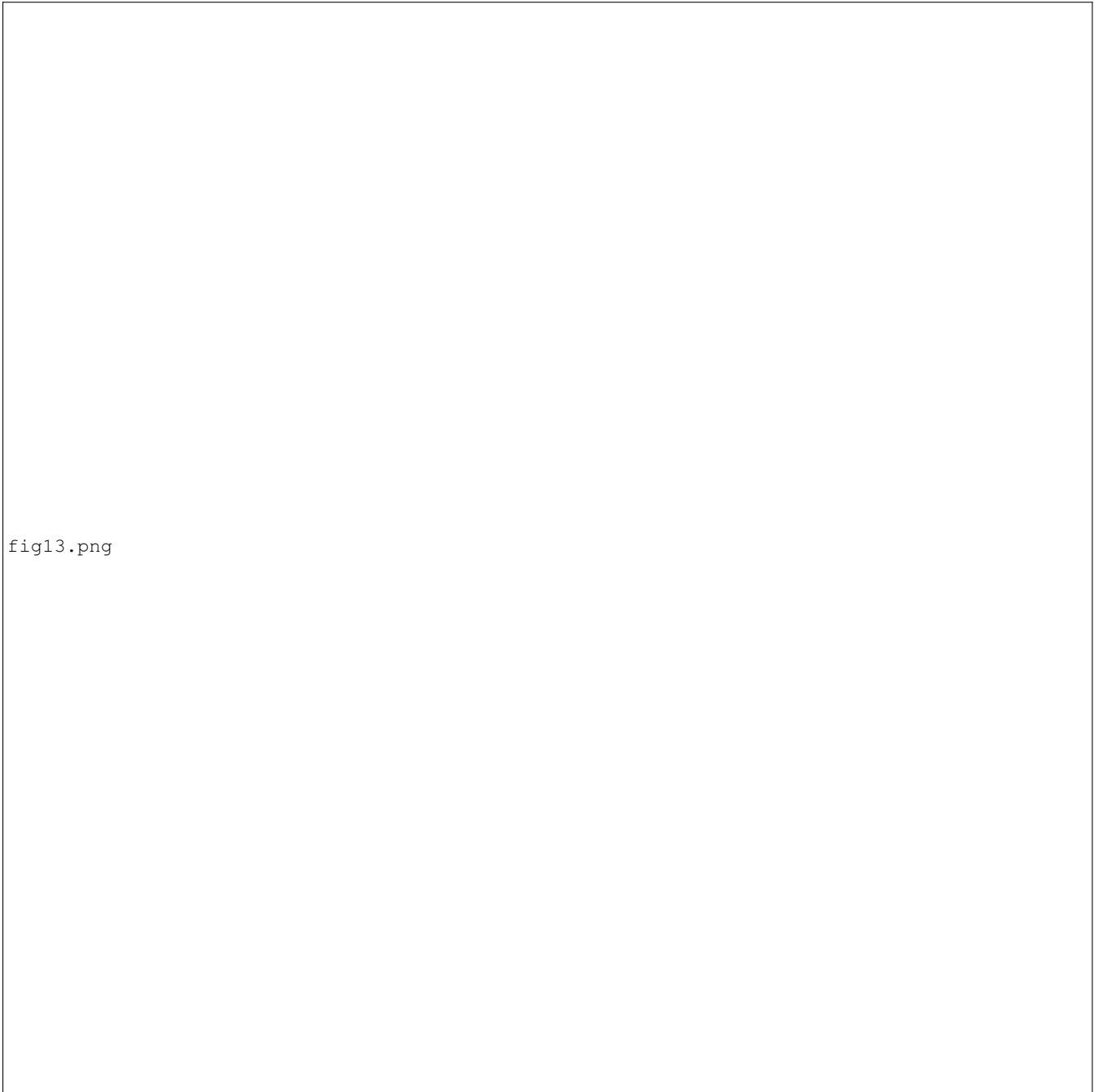
**Figure 13.** Thermal correction comparison ~~from launch~~ showing uncorrected temperature (°C), corrected temperature (°C), and the 00Z Topeka sounding environmental temperature profile (°C).
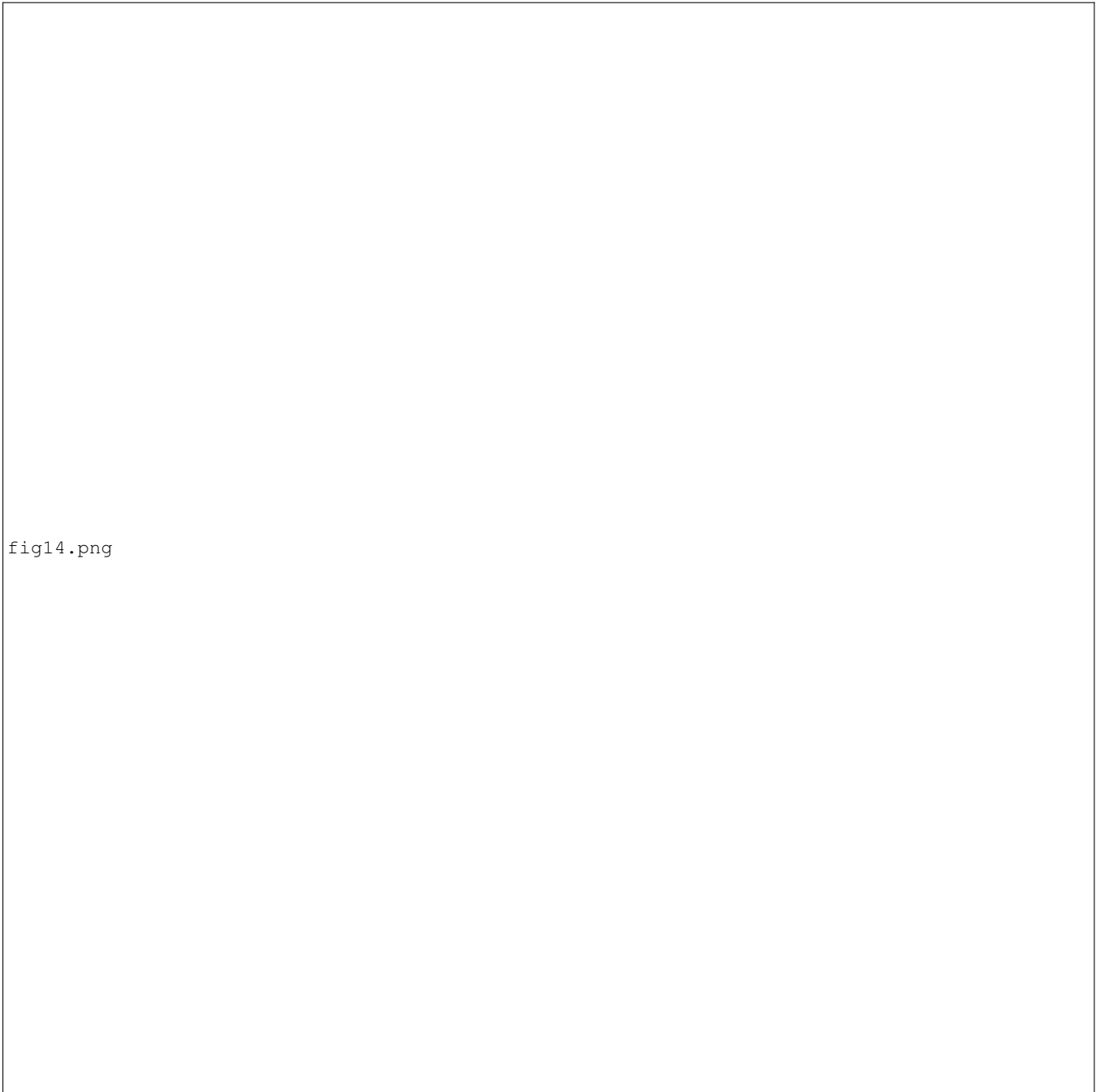
**Figure 14.** GPS data and pressure perturbation sensor data for tornado and updraft portion of the probe flight.